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B.Tech.(Aerospace Engg.) (2012 Onwards) (Sem.-4)

AEROSPACE PROPULSION – I

Subject Code: ASPE-207 M.Code: 71531

Time: 3 Hrs. Max. Marks: 60

INSTRUCTIONS TO CANDIDATES:

- SECTION-A is COMPULSORY consisting of TEN questions carrying TWO marks each.
- SECTION-B contains FIVE questions carrying FIVE marks each and students have to attempt any FOUR questions.
- SECTION-C contains THREE questions carrying TEN marks each and students have to attempt any TWO questions.

SECTION-A

Answer briefly :

- Name two types of gas turbine engines which primarily provides shaft power output.
- Write trust equation for a jet engine which is operating at fully expanded nozzle condition.
- c. How is an isentropic flow different from an adiabatic flow?
- d. Define total pressure.
- e. Which thermodynamic cycle does a gas turbine works on?
- Write the relation for isentropic efficiency of a compressor.
- g. Which thrust augmentation is utilized during take-off in hot weather?
- Name any 4 categories of rocket engine types based on propellants.
- i. What is uninstalled thrust?
- j. What is the maximum achievable exit Mach number in convergent nozzle?

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SECTION-B

- 2) Obtain thrust equation for a rocket engine.
- 3) Mention the property variation across an oblique shock on a neat diagram.
- For isentropic flow, obtain the relation for pressure ratio vs temperature ratio.
- Show that the least pressure ratio required across a nozzle to reach sonic speed is 1.89.
- Briefly explain about any two parameters affecting the takeoff thrust of an engine.

SECTION-C

- Classify aircraft engines based on their operational flight Mach no. and explain their limitations.
- With neat diagram, explain the various configurations of solid propellant structures and state their advantages.
- Explain the over expanded and under expanded rocket nozzle operating conditions with help of a neat diagram.

NOTE: Disclosure of Identity by writing Mobile No. or Making of passing request on any page of Answer Sheet will lead to UMC against the Student.

