

Code No: RT41215

R13**Set No. 1**

IV B.Tech I Semester Supplementary Examinations, February/March - 2018

ANALYSIS OF COMPOSITE STRUCTURE

(Aeronautical Engineering)

Time: 3 hours

Max. Marks: 70

*Question paper consists of Part-A and Part-B**Answer ALL sub questions from Part-A**Answer any THREE questions from Part-B*

PART-A (22 Marks)

1. a) What are the limitations of composite materials? [3]
- b) What is meant by orthogonally isotropic material? Give an example. [4]
- c) Distinguish between micro and macro mechanics approach? [4]
- d) Explain what is a cross ply, symmetric and angle ply laminates? [3]
- e) What are the assumptions used in classical laminate theory? [4]
- f) Write the importance of failure envelopes in structural analysis? [4]

PART-B (3x16 = 48 Marks)

2. a) What are the advantages of composite materials? And explain its area of applications? [8]
- b) Enumerate the classification of composites citing one example in each category [8]
3. a) Describe stress strain relations for a lamina of arbitrary orientation? [8]
- b) Using the mechanics of materials approach, derive the rule of mixture, stating the assumptions made? [8]
4. How to determine the Young's modulus in longitudinal and transverse directions for a lamina using mechanics of material approach? [16]
5. a) Find the compliance matrix for a graphite/epoxy lamina. The material properties are given as $E_1 = 181$ GPa, $E_2 = 10.3$ GPa, $E_3 = 10.3$ GPa, $\nu_{12} = 0.28$, $\nu_{23} = 0.60$, $\nu_{13} = 0.27$, $G_{12} = 7.17$ GPa, $G_{23} = 3.0$ GPa and $G_{31} = 7.00$ GPa. [12]
- b) What are the main factors that are peculiar to laminate strength analysis? [4]
6. a) Derive the expression for stiffness matrix and compliance matrix for an angle ply lamina using generalized Hooke's law. [8]
- b) Compute all terms of [A] [B] and [D] matrices for [+45/-45] lamina with the following lamina properties. $E_1=140$ GPa, $E_2=10$ GPa, $E_6=5$ GPa and ν_{12} is 0.3. Take ply thickness $d=0.125$ mm. [8]
7. Briefly discuss the theory and relevant buckling equations for laminated plates. [16]